

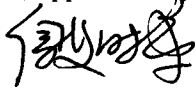
型号认可证数据单

THE VALIDATION DATA SHEET

编号/No: VTC0257A

版次/Revision: 0

型号/Type:
G150

批准人/Approved By:


日期/Date: 2011 年 6 月 21 日

本数据单是型号认可证(编号: VTC0257A)的组成部分，它规定了符合中国民用航空总局的适航要求所颁发此型号认可证的产品的状态和限制。

This data sheet, which is part of Validation of Type Certificate (No: VTC0257A) , prescribes condition and limitation under which the product for which the type certificate was issued meets the airworthiness requirements of the Chinese Civil Aviation Regulation.

型号认可证持有人/Validation of Type Certificate Holder:

Gulfstream Aerospace LP

Department 4199 Ben Gurion International Airport 70100, Israel

有效页清单/List of effective pages :

页数/ Pages	1	2	3	4	5	6	7	8	9	10	11	12	13	14	
版次/Revision	0	0	0	0	0	0	0	0	0	0	0	0	0	0	

第1 部分 概述

SECTION 1 GENERAL

1. CAAC 认可数据单和型号合格证数据单

CAAC Validation Data Sheet and Type Certificate Data Sheet

	中国民用航空总局认可数据 单号、版次和颁发日期 CAAC Validation Data Sheet Number, Revision and issuance date	型号合格证数据单号、版次和 颁发日期 Type Certificate Data Sheet Number, Revision and issuance date
当前 Current	VTC0257A, Revision 0, 2011-6-21	A51L、Rev12, 2007.03.20
历史 History	NONE	NONE

2. 类别/**Category**: 运输类飞机/Transport Airplanes3. 认可当局/**Validation Authority**: 中国民用航空总局/**CAAC**审定当局/**Certifying Authority**: 以色列民航局/**CAAI**4. 型号合格证持有人/**Type Certificate Holder**:

Gulfstream Aerospace LP

Address: Department 4199 Ben Gurion International Airport 70100, Israel

第2 部分 技术特性和使用限制

SECTION 2 Technical Data and Limited

型号 **Gulfstream G150**, 运输类 2005 年 11 月 7 日批准**Model Gulfstream G150, (Transport Category Approved November 7, 2005)****GULFSTREAM G150 型号数据****Data Pertinent to Model GULFSTREAM G150**发动机/**Engines** 两台 Honeywell TFE 731-40AR-200G, FAA 型号合格证数据单 E1NM

2 Honeywell TFE 731-40AR-200G FAA Type Certificate E1NM

燃油 /Fuel	按照批准的 AFM 限制章节,符合 Honeywell 规范的: EMS 53111(型号 Jet A-1&JP-8), EMS53116(型号 JP-5)和 EMS53113(型号 Jet B&JP-4) Conforming to Honeywell Specifications EMS 53111(Jet A type), EMS 53112(Jet A-1&JP-8 types),EMS 53116 (JP-5 type)and EMS 53113 (Jet B & JP-4 tupes) as per LIMITATIONS SECTION of approved AFM No.3 Jet Fuel GB6537-94
滑油/Oil	满足Honeywell 规范 EMS 53110, Type 2. Conforming to Honeywell Specification EMS 53110, Type 2.

发动机限制/Engine Limits

海平面静推力/Static Thrust at Sea Level	4420 磅/ LBS
最大连续/Maximum Continuous	4420
起飞(5 分钟)/Take-off 5 minute)	4420
最大连续允许发动机使用转速 /Maximum Continuous Permissible Engine Operating Speeds for the Engine Rotors, %RPM (RPM)	1 1100.0% (21000) 100.4% (31611) 101.0% (31800)
-低压转子 (N1) / Low pressure rotor (N1)	
-高压转子 (N2) / High pressure rotor (N2) APR Off	
APR On	
最大中级涡轮温度 ITT (°C) Maximum Interstage Turbine Temperature ITT(°C)	
-最大连续,Maximum Continuous	990
- 起飞 (5 分钟) , Take-off (5 minutes)	1004 1022
-起飞 (5分钟) , APR开启 (°C) Take-off (5 minutes) with APR ON	990
开-启期间 During starting	
最大滑油管道温度 /Maximum Oil Inlet Temperature, (°C)	直到 30000 英尺 / Up to 30,000 ft
- 风扇 齿轮 箱 /Fan gearbox	127 157
-2分钟风扇齿轮箱瞬态 / Fan gearbox transient for 2 minutes	149

maximum - 附件 齿 轮 箱 /Accessory gearbox	
滑 油 压 力 (磅 / 平 方 英 寸)/Oil Pressure (PSIG) -正常运行/Normal operating -慢车, 最小/At idle, minimum -最大/Maximum	62-83 50 100
最大引气和功率提取 /Maximum Bleed and Power Extraction: 见Honeywell 安装手册IM-8010 See Honeywell Installation Manual IM-8010	

空速限制/Airspeed Limits:

-Vmo (最大使用 Max operating)	
海平面到 8000 英尺/S.L. to 8000 ft.	310KCAS
从 8000 英尺到 29260 英尺线性增加/increasing linearly from 8000.ft to 12000 ft to	330KCAS
-Mmo 高于 29260 英尺 /Above 29260 ft.	0.85M
-Mmo 高于 24770 英尺, 自动驾驶仪脱开及马赫数配平不工作/Mmo with Autopilot disengaged and Mach Trim inoperative above 24770 ft.	0.78M
-VA (机动性)/(Manoeuvring):	
-VA 从海平面 272KCAS 到 20000 英尺 287KCAS 线性变化/ VA varies linearly from 272KCAS at S.L. to 287 KCAS at 20000ft.	
-VA 从 20000 英尺 287KCAS 到 29300 英尺 330KCAS 线性变化/VA varies linearly from 287 KCAS at 20000 ft. to 330 KCAS at 29300 ft.	
-高于 29300 英尺/Above 29300 ft.	0.85 M
-VFE (襟翼 12° 和缝翼 25° /Flaps 12° and slats 25°)	250 KCAS
-VFE (襟翼 20° 和缝翼 25° /Flaps 20° and slats 25°)	225 KCAS
-VFE(襟翼着陆位 40° 和缝翼 25° /Flaps Landing 40° and slats 25°)	180 KCAS
-VSB (减速板工作/Airbrakes Operation) 无速度限制/No speed limitation	
-VLE and VLO (起落架放下和工作速度/L/G Extension & Operating Speed)	180 KCAS
-VMCA(襟翼 12° 和 20° /Flaps 12° and 20°)	101 KCAS
-VMCG(襟翼 12° 和 20° /Flaps 12° and 20°)	103 KCAS
-轮胎限制的地面速度/Tire limit ground speed : KTS (MPH)	182 KTS (210)

最大运行高度/ Maximum Operating Altitude:

45000 英尺/ft.

重心范围: 批准的重心范围见下表

重心范围/C.G. Range 批准的重心范围见下表 (起落架放下和收上的力矩变化可忽略)

/Approved center of gravity range is as shown in the following tables (Gear extension and retraction moment change is negligible) :

13200 磅/LBS— 37.00 % MAC
 14330 磅/LBS — 20.00 % MAC
 18700 磅/LBS — 20.00 % MAC
 23700 磅/LBS—24.50 % MAC
 26250 磅/ LBS —30.60 % MAC
 26250 磅/ LBS — 38.00 % MAC
 22000 磅/ LBS — 38.00 % MAC
 21700 磅/LBS—37.00 % MAC

基准/Datum:

两数据点之间线性变化/LINEAR VARIATION BETWEEN POINTS

机身 0 站位位于基准点 14L/14R 之前 135.65 英寸, 14L/14R 是位于前机身站位 STA. 135.65, BL $\pm$ 36.99, WL35.00 的一个突出铆钉头

/Fuselage station 0 is located 135.65 inches forward of alignment points 14L / 14R Points
 14L / 14R are the protruding rivets heads located in FWD fuselage at STA. 135.65, BL $\pm$ 36.99, WL 35.00

平均空气动力弦长/Mean Aerodynamic Chord (MAC) :

86.26 英寸, 位于机身站位 317.087 的机翼前缘处。

86.26 inches with leading edge at fuselage station 317.087

纵向调平: 在任一座椅导轨 (机身站位 298.3 (FR 30)) 平行于飞机中心线放置水平仪。

Levelling Longitudinally : Place level on either seat rail at fuselage station 298.3 (FR 30) parallel to A/C centerline.

横向调平: 在任一座椅导轨 (机身站位 298.3 (FR 30)) 垂直于飞机中心线放置水平仪。

Means Laterally : Place level on seat rails at fuselage station 298.3 (FR 30) at 90° to A/C centerline

最大重量-停机坪总重/Maximum Weights -Ramp Gross Weight	26,250 LBS
最大起飞重量/Max Takeoff Weight	26,100 LBS
最大着陆重量/Max Landing Weight	21,700 LBS
最大零燃油重量/Max Zero Fuel Weight	17,500 LBS

最小机组/Minimum Crew 2 (驾驶员和副驾驶/Pilot and Copilot)

最大旅客数/Maximum Passengers: 9, 见注 5/Nine See Note 5

最大行李重量/Maximum Baggage 1100
 磅/LBS 力臂(英寸) /ARM (INCHES) 452

燃油量/Fuel Capacity :

	左/右机翼油箱 LH / RH Wing tanks	集油箱 Collectors	CTS CTS	机身 Fuselage
总/可用燃油 (磅) /Total Usable Fuel (LBS)	3592 / 3574	116 / 105	1317 / 1315	5308 / 5306
力臂(英尺)/ /Arm (INCHES)	342	334.8	310.6	388.2
不可用燃油 (磅) /Unusable Fuel (LBS)	18.6	11	1.6	1.8
力臂/Arm (英尺/INCHES)	342.1	348	310.7	388.4

(燃油系统数据见注 1/See NOTE 1 for data on Fuel System)

滑油量/Oil Capacity :

(密度/Density: 8.2 磅/美加仑/LBS/US Gallon) 二个发动机系统的总量/2 engine systems.

Total (滑油系统数据见注 1/See NOTE 1 for data on Oil System)

总量/TOTAL: 32 可用量/USABLE: 8 力臂/ARM: 496

其它运行限制 飞机应当按照批准的飞机飞行手册中规定的运行限制进行运行 见注 4
 Other Operating Limitations Aircraft shall be operated according to operating limitations specified in approved Airplane Flight Manual see Note 4.

操纵面运动/:	操纵面 Surface	行程 (以后缘为准) Travel (at trailing edge)	容差 Tolerance
副翼/Aileron		向上/Up 15°	± 15'
		向下/Down 15°	± 15'
副翼配平片/Aileron Trim		向上/Up 5°	+0° -1°
		向下/Down 5°	+0° -1°
方向舵/Rudder		左/Left 22°	±30'
		右/Right 22°	±30'
方向舵配平片/Rudder Trim Tab		左/Left 11°30'	+2° -1°
		右/Right 11°30'	+2° -1°
方向舵齿轮片/Rudder Gear Tab		左/Left 7.75°	±1°
		右/Right 7.75°	±1°
升降舵/Elevator		机头向上/Nose Up 22°30'	±30'
		机头向下/Nose Down 12°30'	±15'

升降舵齿轮片/Elevator Gear Tab	机头向上/Nose Up 3°	±30'
	机头向下/Nose Down 18°	±30'
水平安定面调整片/Stabilizer Trim	向上/Up 1°	±15'
(前缘 Leading edge)	向下/Down 12°	±30'
减速板/Airbrakes	向上/Up 45°	±15'
缝翼/Slats	向下/Down 25°	+1-2°
襟翼/Flaps	向下最大/Max Down 40°	+1-1°30'

制造商的 G150 飞机制造序列号自 S/N201 起/Manufacturer's Serial numbers Eligible,
Model GULFSTREAM G150 S/N 201 and subs.

审定基础/Certification Basis

14 CFR part 25,生效日期 1965 年 2 月 1 日,包括修正案 25-1 到 25-108,除以下条款外。

14 CFR part 25, effective February 1, 1965, including amendments 25-1 through 25-108, except for the following paragraphs which are complied with at the identified amendment level.

FAR25	章节 SUB-SECTION	FAR 部分标题 FAR SECTION TITLE	适用区域/系统 ALLPICABLE TO ZONE/SYSTEM	G150 修正案 AMDT
101		性能-总则/ PERFORMANCE - GENERAL		38
105		起飞/TAKEOFF		0
109		加速-停车距离 ACCELERATE-STOP DISTANCE		42
113		起飞距离和起飞滑跑距离 TAKEOFF DISTANCE & TAKEOFF RUN		23
115		起飞飞行航迹 TAKEOFF FLIGHT PATH		0
143	f	总则-控制和机动 CONTROL & MANEUVER - GENERAL		42
149	h	最小操纵速度		72

		MINIMUM CONTROL SPEED		
201	c,d	失速演示 STALL DEMONSTRATION		42
203		失速特征 STALL CHARACTERISTICS		0
253		高速特征 HIGH SPEED CHARACTERISTICS		72
305		强度和形变 STRENGTH AND DEFORMATION	后机身 (*), 水平安定面, 垂尾, 起落架 AFT FUSELAGE (*), HORIZONTAL STABILIZER, VERTICAL TAIL, LANDING GEAR	54
305		强度和形变 STRENGTH AND DEFORMATION	机翼, 吊舱 WING, NACELLE	77
307		结构符合性证明 PROOF OF STRUCTURE	后机身, 水平安定面, 垂尾, 起落架, 机翼, 吊舱 AFT FUSELAGE, HORIZONTAL STABILIZER, VERTICAL TAIL, LANDING GEAR, WING, NACELLE	54
321		飞行载荷: 总则 FLIGHT LOADS - GENERAL	后机身, 吊舱, 垂尾, 机翼 AFT FUSELAGE, NACELLE, VERTICAL TAIL, WING	23
333		飞行包线 FLIGHT ENVELOPE	后机身, 吊舱, 垂尾, 机翼 AFT FUSELAGE, NACELLE, VERTICAL TAIL, WING	0
335		设计空速 DESIGN AIRSPEED	后机身, 吊舱, 垂尾, 机翼 AFT FUSELAGE, NACELLE, VERTICAL TAIL, WING	23
341		突风负载 GUST LOADS	后机身, 吊舱, 垂直尾翼, 机翼 AFT FUSELAGE, NACELLE, VERTICAL TAIL, WING	0
343		设计燃油和滑油载荷 DESIGN FUEL & OIL LOADS	后机身, 吊舱, 垂尾, 水平安定面, 机翼	18

			AFT FUSELAGE, NACELLE, VERTICAL TAIL, HORIZONTAL STABILIZER, WING	
345		增压装置 HIGH LIFT DEVICES		46
349		滚转情况 ROLLING CONDITIONS	吊舱, 机翼 NACELLE, WING	23
351		偏航机动情况 YAWING CONDITIONS	后机身, 吊舱, 垂尾, 水平安定面 AFT FUSELAGE, NACELLE, VERTICAL TAIL, HORIZONTAL STABILIZER	46
361		发动机扭矩 ENGINE TORQUE		46
363		发动机支架侧向载荷 SIDE LOAD ON ENGINE MOUNT		23
365		增压舱载荷 Pressurized compartment load		54
371		陀螺载荷 GYROSCOPIC LOADS		0
373		速度控制装置 SPEED CONTROL DEVICES		0
391		操纵面载荷: 总则 CONTROL SURFACE LOADS- GENERAL	机翼, 垂直尾翼 WING, VERTICAL TAIL	0
395		控制系统 CONTROL SYSTEM	机翼, 垂直尾翼, 飞行控制系统 WING, VERTICAL TAIL, FLIGHT CONTROLS SYS.	23
397		操纵系统载荷 CONTROL SURFACE LOADS	机翼, 垂尾, 水平安定面, 飞行控制系统 WING, VERTICAL TAIL, HORIZONTAL STABILIZER, FLIGHT CONTROL SYS.	38
399		双操纵系统 Dual control system		72
415		地面突风情况 GROUND GUST CONDITIONS	垂尾, 水平安定面, 飞行控制系统 VERTICAL TAIL, HORIZONTAL SABILIZER, FLIGHT CONTROLS SYS.	0
427		非对称载荷 UNSYMETRICAL LOADS	机身尾部, 垂尾 AFT FUSELAGE, VERTICAL TAIL	23

445		辅助气动力面 AUXILLIARY AERODYNAMIC SURFACES	小翼 WINGLETS	0
459		特殊装置 SPECIAL DEVICES	金属板 SLATS	0
473		着路载荷情况和假设 GROUND LOADS CONDITIONS & ASSUMPTIONS	起落架, 吊舱, 机翼, 后机身, 垂尾, 水平安定面 LANDING GEAR, NACELLE, WING, AFT FUSELAGE, VERTICAL TAIL, HORIZONTAL SABILIZER	23
479		水平着陆情况 LEVEL LANDING CONDITIONS	起落架, 吊舱, 机翼, 后机身 LANDING GEAR, NACELLE, WING, AFT FUSELAGE	23
481		尾沉着陆情况 TAIL DOWN LANDING CONDITIONS	起落架, 吊舱, 机翼, 后机身 LANDING GEAR, NACELLE, WING, AFT FUSELAGE	0
483		单起落架着陆情况 ONE WHEEL LANDING CONDITIONS	后机身, 起落架, 吊舱, 机翼 AFT FUSELAGE, LAND GEAR, NACELLE, WING	0
485		侧向载荷情况 SIDE LOAD CONDITIONS	后机身, 起落架, 吊舱, 机翼 LANDING GEAR, NACELLE, WING, AFT FUSELAGE	0
491		滑行、起飞和着陆滑跑 TAKEOFF RUN	后机身, 起落架, 吊舱, 机翼 LANDING GEAR, NACELLE, WING, AFT FUSELAGE	0
493		滑行刹车情况 BRAKED ROLL CONDITIONS	后机身, 起落架 AFT FUSELAGE, LANDING GEAR	23
499		前轮侧偏 NOSE WHEEL YAW	后机身, 起落架 AFT FUSELAGE, LANDING GEAR	46
561		应急着陆情况 EMERG. LANDING CONDITION	后机身, 吊舱 AFT FUSELAGE, NACELLE	23
519		顶升和系留装置 JACKING AND TIE-DOWN PROVISION		0
571		结构损伤容限和疲劳评定 DAMAGE-TOLERANCE AND FATIGUE EVALUATION OF STRUCTURE		54
693		关节接头 JOINTS		0

697		升力和阻力装置及操纵器件 LIFT AND DRAG DEVICES CONTROLS		46
701		襟翼和缝翼的交联 FLAP & SLAT INTERCONNECTION		46
723		减震试验 SHOCK ABSORPTION TESTS		46
725		限制落震试验 LIMIT DROP TESTS		23
727		储备能量吸收落震试验 RESERVE ENERGY ABSORP. DROP TESTS		23
729		收放机构 RETRACTING MECHANISM		42
731		WHEELS 机轮		0
733		TIRES 轮胎		48
735		BRAKES 刹车		48
783	e	DOORS 舱门		54
833		燃烧加温系统 COMBUSTION HEATING SYSTEM		0
855		货舱和行李舱 CARGO OR BAGGAGE COMPARTMENT	AFT FUSELAGE	32
856		隔热隔音材料的安装 THERMAL ACOUSTIC INSULATION MATERIALS		111
857		货舱等级 CARGO COMPT CLASSIFICATION	AFT FUSELAGE	32
951		燃油系统-总则 FUEL SYSTEM - GENERAL		38
979		压力加油系统 PRESSURE FUELING SYSTEM		38
981		燃油箱防火 FUEL TANK (IGNITION PREVENTION)		(**)
1351		电子系统及设备-总则 ELECT. SYSTEMS & EQUIPT. - GENERAL		41
1416		气动除冰系统 PNEUMATIC DE-ICER BOOT SYSTEM		46

1419		防冰 ICE PROTECTION	0
1433		真空系统 VACUUM SYSTEM	0
1435		液压系统 HYDRAULIC SYSTEMS	41

注: (*) 后机身的定义为站位 10824 (43 框)

(**) GALP 选择按照修正案 25-102 的 25.981,除了 25.981(c)外

Notes: (*) Aft fuselage is defined Aft of Station 10824 (frame 43)

(**) GALP elected to comply with 25.981 at amendment 25-102 except for paragraph 25.981(c)

2. 专用技术条件 Special Conditions:

FAA 25 –ANM-104,高强度辐射场

FAA 25 –ANM-104, High Intensity Radiated Fields (HIRF)

3. 等效安全水平/Equivalent Level of Safety Findings:

Section 25.812 (a)(1), (c), (d), (e) 应急照明

Section 25.812 (a)(1), (c), (d), (e) Emergency Lighting

Section 25.841 (b) (6) 客舱增压—高原起飞和着陆运行

Section 25.841 (b) (6) Cabin Pressurization – High Altitude Take-off and Landing Operations

Section 25.1121(c) 在排气系统部件的防火分离

Section 25.1121(c) Exhaust system components separated by fireproof shields

Section 25.1203 (a) 涡轮排气管区域的火警探测系统

Section 25.1203 (a) Fire Detection System within the Turbine Tailpipe Zone

Section 25.1305 (c) (3) 仅用数字显示器显示发动机高压涡轮转子转速

Section 25.1305 (c) (3) Digital Display Only of Turbine Engine High Pressure Rotor Speed (N2)

Section 25.255(a) 失配平条件下的机动稳定性和操纵性

Section 25.255(a) Maneuvering stability and controllability in a mistrim condition

4. 豁免/Exemptions

Section 25.901 (c) 不可控发动机高推力

Section 25.901 (c) Uncontrollable High Engine Thrust

5. 自愿符合的条款/Optional Design Regulations:

自愿符合下列条款:

Compliance with the following optional requirements has been established:

Section 25.801 水上迫降/Ditching

Section 25.1419 防冰 Ice Protection

6. 噪音标准/Noise Standards

14 CFR part 36,生效日期 1969 年 12 月 1 日, 包括修正案 36-1 到 36-26 (Stage 4)

14 CFR part 36, effective December 1, 1969, including amendments 36-1 through 36-26
(Stage 4)

7. 燃油排气和废弃物排放标准/ Fuel Venting and Exhaust Emissions Standards

14 CFR part 34,生效日期 1990 年 9 月 10 日, 包括修正案 34-3

14 CFR part 34, effective September 10, 1990 including amendment 34-3

生产基础/Production Basis. Production Certificate No. PA-30.

设备/Equipment.

按照适用的适航条款（见审定基础）所要求的基本设备必须安装在飞机上。

The basic required equipment as prescribed in the applicable airworthiness regulation (see certification basis) must be installed in the aircraft for certification.

G150 的主设备清单报告 NO.25G000/051724。

Model GULFSTREAM G150—Master Equipment List Report
NO.25G000/051724

注: /NOTES:

注 1:/NOTE1:

(a)每架飞机必须提供在初始合格审定时审定的空机重量所包含的设备清单和装载说明的最新重量和平衡报告。

/(a) Current weight and balance report including list of equipment included in certificated empty weight and loading instructions must be provided for each aircraft at the time of original certification.

(b)任何时候飞机装载必须保证重心在规定的限制范围内。

/(b) The airplane must be loaded so that the C.G. is within the specified limits at all times.

(c)下面定义的燃油和滑油系统的液体重量必须包括在空机重量中。

/(c) The weight of fuel and oil systems fluid as defined below must be included in the empty weight of the airplane.

燃油系统-不可用油（油箱，排液装置和管路放油）， 磅（加仑）/FUEL SYSTEM- Unusable (drainable from tanks drain and lines),LBS (Gallons) 不能放油（油箱和管路收集油），磅（加仑） /Undrainable (trapped in tanks and lines), LBS (Gallons)	磅 LBS 22.7(3.4)	ARM (英寸 inch) 343.8
	22.4 (3.3)	394.2
滑油系统无法排空（系统）-总计/OIL SYSTEM Unusable drainable (systems) – Total	磅/LBS 16	ARM (英寸inch) 496

不能排油（系统）-总计 /Undrainable (systems) – Total	8	496
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注 2:NOTE 2.

飞机飞行手册限制区域列出的全部所有要求的标牌必须安装在飞机上的适当位置。/All required placards listed in the Limitations Section of the Airplane Flight Manual must be installed in their proper locations.

注 3:NOTE 3.

(a) 飞机的相应服务和维修的基本信息包含在 AMM 手册 AMM P/N G150-1001-3。

/ (a) Information essential to the proper servicing and maintenance of the aircraft is contained in the Aircraft Maintenance Manual AMM P/N G150-1001-3 .

(b) 强制更换的时间、结构检查间隔和相关检查程序及审定维修要求已纳入经批准的 AMM 手册的适航限制章节（第 05-10-10）中。

/ (b) Mandatory replacement times, structural inspection intervals and related structural inspection procedures and Certification Maintenance Requirements are presented in the approved Airworthiness Limitation Section 05-10-10 of the AMM.

注 4:NOTE 4.

G150 必须按照经批准的飞行手册 AFM P/N G150-1001-1 操作。

/Airplane Model GULFSTREAM G150 must be operated according to the Approved Airplane Flight Manual marked P/N G150-1001-1.

注 5:NOTE 5.

本型号合格证定义不包括飞机的外部喷漆和内部布局及旅客设施。依据合格审定基础，本型飞机具备达到 9 名旅客的座位布局和设施的能力。

/This Aircraft Type Certificate defines a configuration which does not include exterior painting of the aircraft and its interior furnishing and passenger provision. The Aircraft is eligible for carriage of up to 9 passengers provided approved seating arrangement and related required passenger provisions are incorporated in accordance with the Basis of Certification.

注 6:NOTE 6.

根据批准的型号设计，执行了 MOD G15-10015 改装的 G150 飞机的 RVSM 能力符合 FAR 91 附录 G 第 2 节的技术要求。但是 RVSM 的运营还必须由获得局方运行合格审定部门批准。

/Per the Approved Type Design, Model G150 complies with the RVSM technical requirements contained in FAR 91 Appendix G Section 2, provided MOD G15-10015 is implemented. However, operational approval to fly in RVSM airspace must still be granted by the CAAI Operation Branch.

..... END.