

型号认可证数据单

THE VALIDATION DATA SHEET

编号/No.: VTC0249E

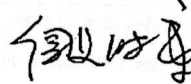
版次/Revision: 2

型号/Type:

AS907-1-1A, AS907-2-1G,

AS907-2-1A, AS907-3-1E

批准人/Approved by:



日期/Date: 2014 年 12 月 29 日

本数据单是型号认可证（编号 VTC0249E）的组成部分，它规定了符合中国民用航空局的适航要求所颁发此型号认可证的产品的状态和限制。

This data sheet, which is a part of Validation Type Certificate (No: VTC0249E), prescribes conditions and limitations under which the product for which the Type Certificate was issued meets the airworthiness requirements of the Chinese Civil Aviation Regulation.

型号认可证持有人/Validation of Type Certificate Holder:

Honeywell International Inc.

111 South 34th Street

Phoenix, AZ, 85034

有效页清单/List of effective pages :

页数/ Pages	1	2	3	4	5	6	7	8	9	10	11
版次/Revision	2	2	2	2	2	2	2	2	2	2	2

第1部分 概述

SECTION 1 GENERAL

1. CAAC认可数据单和型号合格证数据单

CAAC Validation Data Sheet and Type Certificate Data Sheet

	中国民用航空局认可数据单号、版次和颁发日期 CAAC Validation Data Sheet Number, Revision and issuance date	型号合格证数据单号、版次和颁发日期 Type Certificate Data Sheet Number, Revision and issuance date
当前 Current	No.VTC0249E, 版次 2, 2014 年 12 月 29 日颁发 No.VTC0249E; Revision 2; Date of Issue: 2014-12-29	No. E00010LA, 版本: 6, 2014 年 9 月 15 日颁发 No. E00010LA, Revision 6, Date of Issue: 2014-9-15
历史 History	No.VTC0249E, 版次 1, 2013 年 5 月 15 日颁发 No.VTC0249E; Revision 1; Date of Issue: 2013-5-15	No. E00010LA, 版本: 3, 2011 年 3 月 18 日颁发 No. E00010LA, Revision 3, Date of Issue: 2011-3-18
	No.VTC0249E, 版次 0, 2010 年 12 月 2 日颁发 No.VTC0249E; Revision 0; Date of Issue: 2010-12-2	No. E00010LA, 版本: 2, 2004 年 4 月 14 日颁发 No. E00010LA, Revision 2, Date of Issue: 2004-4-14

2. 类别/Category: 航空发动机/Aeroengine
3. 认可当局/Validation Authority: 中国民用航空局/CAAC
审定当局/Certifying Authority: 美国联邦航空局/FAA
4. 型号合格证持有人/Type Certificate Holder: Honeywell international Inc.
111 South 34th Street
Phoenix, AZ, US 85034
6. ETOPS: 不适用/NA
7. CAAC 专用要求/Special Requirements: 满足中国 3 号燃油 (GB6537-2006) 的要求。
/Compliance with requirement of P.R.China JET Fuel
No.3 (GB6537-2006).

第2部分

SECTION 2

型号/MODELS	AS907-1-1A	AS907-2-1A	AS907-2-1G	AS907-3-1E
类型/TYPE	涡扇：一级风扇，四级轴流高压压气机和一级离心高压压气机，环形燃烧室，两级高压涡轮，三级低压涡轮。 Turbofan: One stage fan, four stage axial flow high pressure compressor, one stage centrifugal high pressure compressor, annular combustor, two stage high pressure turbine, and three stage low pressure turbine.			
净推力额定值（见注 4 和注 12）/Static Thrust Ratings (see NOTE 4 and NOTE 12)	海平面最大连续，磅 /Max Continuous at Ambient Temperature, at Sea Level, lbs 6929 at ISA +15°C	海平面最大连续，磅 /Max Continuous at Ambient Temperature, at Sea Level, lbs 7530 at ISA +14°C	海平面最大连续，磅 /Max Continuous at Ambient Temperature, at Sea Level, lbs 7337 at ISA +17°C	海平面最大连续，磅 /Max Continuous at Ambient Temperature, at Sea Level, lbs 7335 at ISA +22.5°C
	海平面起飞，磅（见注 10）/Takeoff, at Ambient Temperature, at Sea Level, lbs (see NOTE 10) 6944 at ISA +20°C	海平面起飞，磅（见注 10）/Takeoff, at Ambient Temperature, at Sea Level, lbs (see NOTE 10) 7530 at ISA +20°C	海平面起飞，磅（见注 10）/Takeoff, at Ambient Temperature, at Sea Level, lbs (see NOTE 10) 7765 at ISA +17°C	海平面起飞，磅（见注 10）/Takeoff, at Ambient Temperature, at Sea Level, lbs (see NOTE 10) 7638 at ISA +18°C
控制/Controls	燃油控制和功率管理由全权限数字电子控制系统完成，该系统带有两个电子控制单元，双通道控制。系统的硬软件和液压机械组件、整体燃油泵由批准的发动机设备清单控制（见注 13） Fuel control and power management are controlled by a Full Authority Digital Electronic Control (FADEC) system which features a dual-channel electronic control in the form of two electronic control units (ECUs). The hardware and software configurations of this system and the associated engine hydromechanical unit with integral fuel pump are controlled by an approved engine equipment list. (See NOTE 13).			
主要尺寸/Principal Dimensions	尺寸和重心位置参见发动机安装图纸 Refer to the engine installation drawing for dimensions and center of gravity location.			
干质量，磅（最大）/Weight, Dry, Pounds (maximum)	1534	1534	1534	1514
燃油/Fuel	遵从 Honeywell 公司 EMS53111 (A 型燃油)、EMS53112(A-1 和 JP-8 型燃油)、EMS53113 (Jet B 和 JP-4 型燃油)和 EMS53116 (JP-5 型燃油)规范。			

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型号/MODELS	AS907-1-1A	AS907-2-1A	AS907-2-1G	AS907-3-1E
	Fuels conforming to Honeywell International Inc. Specifications EMS53111 (Jet A Type), EMS53112 (Jet A-1 and JP-8 Types, EMS53113 (Jet B and JP-4 Types) and EMS53116 (JP-5 Types). 当从主燃油到替换燃油转换时, 不需要燃油控制调节 No fuel control adjustment is required when changing from primary to alternate fuels. 批准的燃油牌号和添加剂参见发动机安装手册 (见注 12) Refer to engine installation manual for approved fuel types and fuel additives (see NOTE 12)			
滑油/Oil	遵从霍尼韦尔公司 EMS53110 规范 II 类。批准的滑油参见发动机安装手册 (见注 12)。 Oil conforming to Honeywell International Inc. Specification EMS53110, Type II. Refer to engine installation manual for approved engine oils (see NOTE 12)			

审定基础/CERTIFICATION BASIS

AS907-1-1A AS907-2-1G	1965 年 2 月 1 日生效的 14 CFR 第 33 部, 包含 2000 年 12 月 13 日生效的第 1 到 20 修正案。/ 14 CFR Part 33, effective February 1, 1965, as amended by 33-1 through 33-20, effective December 13, 2000. 1990 年 9 月 10 日生效的 14 CFR 第 34 部, 包含 2013 年 10 月 23 日生效的第 1 到 5A 修正案 (见注 15)。/ 14 CFR Part 33, effective September 10, 1990 as amended by 34-1 through 34-5A, effective October 23, 2013. Fuel Venting and Exhaust Emissions Standards (see NOTE 15);
AS907-2-1A	1965 年 2 月 1 日生效的 14 CFR 第 33 部, 包含 2012 年 9 月 20 日生效的第 1 到 33 修正案, 不包含第 22 和 31 修正案。/ 14 CFR Part 33, effective February 1, 1965, as amended by 33-1 through 33-33, effective September 20, 2012, excluding Amendments 33-22 and 33-31. 1990 年 9 月 10 日生效的 14 CFR 第 34 部, 包含 2013 年 10 月 23 日生效的第 1 到 5A 修正案 (见注 15)。/ 14 CFR Part 33, effective September 10, 1990 as amended by 34-1 through 34-5A, effective October 23, 2013. Fuel Venting and Exhaust Emissions Standards (see NOTE 15);
AS907-3-1E	1965 年 2 月 1 日生效的 14 CFR 第 33 部, 包含 2012 年 9 月 20 日生效的第 1 到 33 修正案, 不包含第 22 修正案。/ 14 CFR Part 33, effective February 1, 1965, as amended by 33-1 through 33-33, effective September 20, 2012, excluding Amendments 33-22. 1990 年 9 月 10 日生效的 14 CFR 第 34 部, 包含 2013 年 10 月 23 日生效的第 1 到 5A 修正案 (见注 15)。/ 14 CFR Part 33, effective September 10, 1990 as amended by 34-1 through 34-5A, effective October 23, 2013. Fuel Venting and Exhaust Emissions Standards (see NOTE 15);

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发动机型号列表/Engine Table of Models

型号 Model	申请日期 Date of Application	型号合格证颁发日期 Type Certificate Issued	首次生产日期 1st Production Date
AS907-1-1A	1998 年 11 月 4 日 November 4, 1998	2002 年 6 月 25 日 June 25, 2002	2003 年 5 月 29 日 May 29, 2003
AS907-2-1G	2006 年 4 月 20 日 April 20, 2006	2011 年 3 月 18 日 March 18, 2011	2011 年 3 月 25 日 March 25, 2011
AS907-2-1A	2012 年 8 月 31 日 August 31, 2012	2014 年 5 月 13 日 May 13, 2014	2014 年 5 月 14 日 May 14, 2014
AS907-3-1E	2009 年 2 月 26 日 February 26, 2009	2014 年 7 月 7 日 July 7, 2014	2014 年 7 月 12 日 July 12, 2014

生产基础/ 1965 年 3 月 4 日颁发的 413 号生产许可证, 2000 年 1 月 25 日颁发 413NM 生产许可证
PRODUCTION BASIS /Production Certificate No. 413, issued March 4, 1965. Reissued Production Certificate No. 413NM to Honeywell International Inc. on January 25, 2000.

注释/ NOTE

注 1./NOTE 1. 发动机高压转子和低压转子的最大允许转速/Maximum permissible operating speeds for the low- and high-pressure rotors of the engine are as follows:

型号 Model	低压转子转速/Low-Pressure Rotor Speed (N ₁), RPM			高压转子转速/High-Pressure Rotor Speed (N ₂), RPM		
	最大连续 /Maximum Continuous	起飞(见注 10) /Takeoff (see NOTE 10)	最大瞬态 (20 秒最大状态) /Maximum Transient (20 Seconds Maximum)	最大连续 /Maximum Continuous	起飞(见注 10) /Takeoff (see NOTE 10)	最大瞬态 (20 秒最大状态) /Maximum Transient (20 Seconds Maximum)
AS907-1-1A	9723	9812	9957	27319	27568	28075
AS907-2-1A	9800	9830	9957	27530	27714	28075
AS907-2-1G	9800	9830	9957	27530	27686	28075
AS907-3-1E	9830	9830	9957	27530	27686	28075

注 2./NOTE 2. 温度限制/Temperature Limits:
 最大指示涡轮级间温度/Maximum Indicated Interturbine Temperature (ITT) Limits: °F (°C)

型号 Model	最大连续 Max. Continuous		起飞 (见注 10) Takeoff (see NOTE 10)		起动 (地面/空中) Starting (Ground/Air)	瞬态 Transient

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AS907-1-1A	1702 (928)	1735 (946)	*	1764 (962) **
AS907-2-1A	1742 (950)	1751 (955)	*	1778 (970) **
AS907-2-1G	1742 (950)	1751 (955)	*	1778 (970) **
AS907-3-1E	1742 (950)	1751 (955)	*	1778 (970) **

* 随 N2 转速变化, 参见发动机安装手册 (见注 12) /Varies with N2 speed, refer to engine installation manual (see NOTE 12)

** 20 秒最大/20 seconds maximum

*** 2 分钟最大/2 minutes maximum

最大滑油进口温度/Maximum Oil Inlet Temperature Limits:

连续/Continuous, °F(°C)	瞬态 (2 分钟最大) /Transient (2 Minutes Max.), °F(°C)
280 (138)	310 (154)

外部发动机零件最大温度 (特别零件的温度限制) 已在发动机安装手册中指定 (见注 12) /External engine components maximum temperature (limiting temperature of specific components) is as specified in the engine installation manual (see NOTE 12).

启动期间, 对 AS907-1-1A、AS907-2-1A、AS907-2-1G, 发动机燃油进口温度在 185°F(85°C)、对 AS907-3-1E, 发动机燃油进口温度在 131°F(55°C), 蒸汽体积与液体体积比为 0.45, 到燃油进口温度在 -65°F (-54°C)、粘度 12 厘沱或更低时, 均在发动机安装手册中得到批准 (见注 12), Operation at an engine fuel inlet temperature as high as 185°F(85°C) for the AS907-1-1A, AS907-2-1A and AS907-2-1G engines and 131°F(55°C) for the AS907-3-1E engine with a vapor volume to liquid volume ratio (V/L) equal to 0.45, and as low as -65°F (-54°C)°F with fuel at a viscosity of 12.0 centistokes or less during starting is approved as specified in the engine installation manual (see NOTE 12).

注 3./NOTE 3.

燃油和滑油压力限制/Fuel and Oil Pressure limits:

燃油泵进口压力/Fuel pump inlet pressure:

最小值-下列数值中的最大值/Minimum – Whichever is highest of the following:

- 大于真蒸汽压力 5 psi/5 psi above the true vapor pressure of the fuel;
- 对应蒸汽体积与液体体积比为 0.45 的压力/Pressure corresponding to a V/L ratio of 0.45;
- 35%大气压力/35% of atmospheric pressure;
- 2 psia.

最大值:

- 35 psig

滑油压力/Oil pressure:

滑油压力不受调节, 随 N2 转速变化。参见发动机安装手册 (见注 12) /Oil pressure is not regulated and pressure limits vary with N2. Refer to engine installation manual (NOTE 12).

注 4./NOTE 4.

额定值基于静态试验, 并在下述条件下得到/The ratings are based on static test stand operation

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and under the following conditions:

- a) 没有航空器附件传动载荷/ No loading of aircraft accessory drives;
- b) 没有航空器压气机引气提取/ No aircraft compressor bleed air extraction;
- c) 风扇排气和涡轮排气喷嘴遵照霍尼韦尔 N10780-1 和 N10781-1 图纸/Fan exhaust and turbine exhaust nozzles conforming to Honeywell International Inc. Drawings N10780-1 and N10781-1;
- d) 钟形进口遵照霍尼韦尔 5837800-1 图纸/Bellmouth inlet conforming to Honeywell International Inc. Drawing 5837800-1;
- e) 干进口空气/Dry inlet air;
- f) 无排气喷管反压/No exhaust nozzle back pressure;
- g) 静态推力额定值转点外界温度/Static Thrust Ratings ambient temperature at breakpoint.

注 5./NOTE 5.

附件传动/Accessory Drive Provisions:

附件传动 /Accessory Drive	传动类型(每种 一类)/Drive Type (One Each)	内花键构造 /Internal Spline Configuration	转速和面对驱动 端的转向/RPM and Rotation Facing Drive End	注 (b) /Note (b) 附件最大扭矩 (磅- 英寸) /Accessory Max Torque (lb-in)			最大重量 (磅) /Maximum Weight (pounds)	悬挂力矩 (磅-英 寸) /Overhang Moment (lb-in)**
				Tc	To	Ts		
发电机/交流 发电机 Generator/Alt ernator D30* AS907-1-1A AS907-2-1A AS907-2-1G	经修改的 AS468B-AV1: 转速、扭矩、附 件重量和力矩 AS468B-AV1 modified as follows: RPM, torques, accessory weight and moment as shown	AS468B	13665 注 (a) /Note (a) CW	242	363	1600	34.7	128.5
发电机/交流 发电机 Generator/Alt ernator D30*	经修改的 AS468B-AV1: 转速、扭矩、附 件重量和力矩: AS468B-AV1 modified as follows: RPM, torques, accessory weight and moment as shown	AS468B	13665 注 (a) /Note (a) CW	242	363	1600	40.7	190

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液压泵 /Hydraulic Pump D10* AS907-1-1A AS907-2-1A AS907-2-1G	经修改的 AS468B-AV1: 转速、扭矩、附件重量和扭矩 AS468B-AV1 modified as follows: RPM, torques, accessory weight and moment as shown	AS468B	5974 注 (a) /Note (a) CW	250	375	1544	22.3	103.9
液压泵 /Hydraulic Pump D10* AS907-3-1E	经修改的 AS468B-AV1: 转速、扭矩、附件重量和扭矩 AS468B-AV1 modified as follows: RPM, torques, accessory weight and moment as shown	AS468B	5974 注 (a) /Note (a) CW	250	388	1544	22.3	103.9

CW=顺时针/Clockwise
 Tc=连续扭矩/Continuous Torque
 To=扭矩过载 (5 分钟每 4 小时) /Torque Overload (5 min. per 4 hr. period)
 TS=静止扭矩/Static Torque
 * 安装图纸上附件衬垫相同/Accessory pads are identified by these symbols on the installation drawing.
 ** 在快速安装/拆卸界面上./At quick attach/detach (QAD) interface
 注/Notes:
 (a) 驱动转速基于 100%设计 N2 转速, 28100 rpm /Drive speeds are based on a 100% design N2 of 28,100 rpm.
 (b) 总的附件功率提取限制在发动机安装手册中规定 (见注 12) /Total combined accessory power extraction limits are specified in the engine installation manual (see NOTE 12).

注 6./NOTE 6.

用于航空器的引气流量限制, 参见发动机安装手册 (见注 12) /For aircraft bleed airflow limits, refer to the engine installation manual (see NOTE 12).

注 7./NOTE 7.

对于 AS907-1-1A, AS907-2-1A, AS907-2-1G 发动机, 来自发动机泵的燃油被提取用来驱动航空器燃油系统的涡轮泵。参见发动机安装手册 (见注 12) / For the AS907-1-1A, AS907-2-1A, AS907-2-1G engine aircraft installations, fuel from the engine pump is extracted to drive jet or turbine pumps in the aircraft fuel system (motive flow). Refer to the engine installation manual (see NOTE 12).

	对于 AS907-3-1A 发动机, 发动机燃油泵不驱动飞机燃油系统/The AS907-3-1E engine fuel pump does not drive aircraft system motive flow.
注 8./NOTE 8.	发动机满足在 FAR 25 部附件 C 结冰条件下的运行要求/ The engine meets FAA requirements for operation in icing conditions within the envelope defined in Title 14 CFR part 25, Appendix C.
注 9./NOTE 9.	某些发动机零件具有有限寿命。这些限制在发动机小修手册中的适航限制部分规定/Certain engine parts are life-limited. These limits are included in the engine Light Maintenance Manual, Airworthiness Limitations Section.
注 10./NOTE 10.	当出现发动机意外事故时, 5 分钟起飞时间限制可以延长至 10 分钟/ The normal 5-minute takeoff time limit may be extended to 10 minutes for engine out contingency.
注 11./NOTE 11.	功率设定、功率检查和发动机推力输出控制基于低压转子转速(N ₁)。因此, 在发动机组件中, 包含转速传感器/Power setting, power checks and control of engine thrust output in all operations are based on N ₁ . Speed sensors are included in the engine assembly for this purpose.
注 12./NOTE 12.	对于额外的性能, 批准的运行和安装信息, 参见 FAA 批准的发动机安装手册/For additional performance, authorized operation and installation detailed information, refer to FAA approved sections of the engine installation manual as follows: 型号/Model AS907-1-1A: 24-IM-8014 型号/Model AS907-2-1A: 24-IM-8014 型号/Model AS907-2-1G: 24-IM-8029 型号/Model AS907-3-1E: 24-IM-8030
注 13./NOTE 13.	发动机控制系统已批准限时放行(TLD)。关于最大批准放行间隔和发动机控制系统维修要求的适航限制在发动机小修手册中规定/The engine control systems has been approved for Time Limited Dispatch (TLD) operations, and the airworthiness limitations pertaining to the maximum approved dispatch intervals and maintenance requirements of the engine control system are specified in the engine Light Maintenance Manual, Airworthiness Limitations Section.
注 14./NOTE 14.	推荐的发动机检查间隔包含在发动机小修手册第 5 章中/Recommended engine inspection intervals are included in the engine Light Maintenance Manual, Chapter 5.
注 15./NOTE 15.	发动机型号列表中所列的型号均满足 2013 年 10 月 23 日生效的 14 CFR 34 部第 34-5A 修正案和 2012 年 7 月 18 日生效的 40 CFR 87 部所颁布的排放标准。 The following emissions standards promulgated in 14 CFR part 34, Amendment 34-5A, effective October 23, 2013, and 40 CFR part 87, effective July 18, 2012, have been complied with for all models listed in Engine Table of Models. • 发动机燃油排泄标准: 14 CFR §§34.10 (b) 和 34.11; 以及修订的 40 CFR §§87.10 (b) 和 87.11/Fuel Venting Emission Standards: 14 CFR §§34.10(b) and 34.11; in addition, 40 CFR §§87.10(b) and 87.11 as amended. • 冒烟指数 (SN) 排放标准: 14 CFR §34.21 (c) (2); 以及 40 CFR §87.23 (c) (1) /Smoke Number (SN) Emission Standards: 14 CFR §34.21(e)(2); addition, 40 CFR §87.23(c)(1).

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- 一氧化碳 (CO) 排放标准: 14 CFR §34.21 (d) (1) (ii); 以及 40 CFR §87.23 (c) (1)/Carbon Monoxide (CO) Emission Standards: 14 CFR §34.21(d)(1)(ii); in addition, 40 CFR §87.23(c)(1).
- 碳氢化合物 (HC) 排放标准: 14 CFR §34.21 (d) (1) (i); 以及 40 CFR §87.23 (c) (1) /Hydrocarbons (HC) Emission Standards: 14 CFR §34.21(d)(1)(i); in addition, 40 CFR §87.23(c)(1).
- 氮氧化物(NOx)排放标准: 14 CFR §34.23 (a) (2) 和 (b) (1); 以及 40 CFR §87.23 (c) (2) 和 (c) (3) /Oxides of Nitrogen (NOx) Emission Standards: 14 CFR §34.23(a)(2) and (b)(1); in addition, 40 CFR §87.23(c)(2) and (c)(3).

除以型号合格证数据单定义的取证要求为基础的 FAA 符合性证据外, 发动机制造商声明 ICAO 附件 16 卷 II 第三版第 III 部分第 2 章 2.2.2 节对冒烟指数, 2.3.节对一氧化碳和碳氢化合物, 2.3.2.d.ii 节对氮氧化物 (也称为 CAEP/6) 和 2.3.2.e 节对氮氧化物 (也称为 CAEP/8) 和第 II 部分第 2 章对燃油排泄的排放标准也经过了验证。

In addition to the FAA's finding of compliance based on the certification requirements defined in this TCDS, the engine manufacturer has declared that the ICAO emissions standards identified in Annex 16, Volume II, Third Edition, Part III, Chapter 2, Section 2.2.2 for SN, Section 2.3.2 for CO and HC, Section 2.3.2.d.ii for NOx (also known as CAEP/6) and Section 2.3.2.e for NOx (also known as CAEP/8) and Part II Chapter 2 for fuel venting have also been demonstrated.

注 16/NOTE 16. 模型描述-相似性, 差异和特殊特性/Model description – similarities, differences, and special characteristics:

型号/Model	型号描述/Model Description
AS907-1-1A	一级风扇, 四级轴流高压压气机, 一级离心高压压气机, 环形燃烧室, 两级高压涡轮, 三级低压涡轮。 One stage fan, four stage axial flow high-pressure compressor, one stage centrifugal high-pressure compressor, annular combustor, two stage high-pressure turbine, and three stage low-pressure turbine.
AS907-2-1G	除提高了推力额定值、涡轮级间温度和转子速度限制值、更改了离心压气机和高/低压涡轮喷嘴、改进了燃烧系统、客舱风扇引气系统、采用了新的排气混合系统、并调整了 FADEC 系统软件外, 与 AS907-1-1A 相同。 Same as AS907-1-1A except increased thrust ratings and ITT and rotor speed limits. Modified centrifugal compressor and high- and low-pressure turbine nozzles, improved combustion system, provisions for customer fan bleed, a new exhaust mixer nozzle system, and modified FADEC system software.
AS907-2-1A	除提高了最大连续推力额定值和起飞 N2 转速限制值、可选安装界面零件和升级的 ECU 及其软件外, 与 AS907-2-1G 相同。 Same as AS907-2-1G except increased maximum continuous thrust rating and takeoff N2 rotor speed limit, select installation interface parts, and updated ECU and software.
AS907-3-1E	除更改了推力额定值和涡轮级间温度瞬态时间及转子转速限制值、更改风扇进口安装、更改了风扇出口外涵结构和后安装系统、不可调的第一级压气机静子叶片, 更改的压气机和燃烧室机匣、增加附件传动限制值、更改了发电机快速安装、更改了空气涡轮启动机和更改了 FADEC 系统软件外, 与 AS907-2-1A 相同。 Same as AS907-2-1A except modified thrust ratings and ITT transient time and rotor speed limits, modified fan inlet housing, modified fan outer bypass duct and aft mounting system, non-variable stage 1 compressor vanes, modified compressor and combustor cases, increased accessory drive limits, modified

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	generator quick attach-detach (QAD), modified air turbine starter, and modified FADEC system software.
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注 17./NOTE 17.

发动机型号设计不包括反推装置 (T/R)。对于反推装置的安装考虑参见发动机安装手册 (见注 12)。发动机与下列反推装置兼容:

The engine type design does not include a thrust reverser (T/R). Considerations for the installation of a thrust reverser are contained in the engine installation manual (see NOTE 12). The engines have demonstrated compatibility with the following thrust reversers:

型号/Model	反推装置制造商 T/R Manufacturer	反推装置件号 T/R Part number	
		左手/Left Hand	右手/Right Hand
AS907-1-1A	Aircelle (以前是 Hurel-Hispano) Aircell (formerly Hurel-Hispano)	按照反推装置安装图纸 13A025-03-0G Per T/R installation drawing 13A025-03-0G	按照反推装置安装图纸 13A026-02-0G Per T/R installation drawing 13A026-02-0G
		按照反推装置安装图纸 13A016-00-0G Per T/R installation drawing 13A016-00-0G	按照反推装置安装图纸 13A017-00-0G Per T/R installation drawing 13A017-00-0G
AS907-2-1G	Aircell	按照反推装置安装图纸 31A516-01-0G Per T/R installation drawing 31A516-01-0G	按照反推装置安装图纸 31A517-01-0G Per T/R installation drawing 31A517-01-0G
AS907-2-1A	Aircell	按照反推装置安装图纸 33A016-01-0G Per T/R installation drawing 33A016-01-0G	按照反推装置安装图纸 33A017-01-0G Per T/R installation drawing 33A017-01-0G
AS907-3-1E	Aircell	按照反推装置安装图纸 32A716-02-0G Per T/R installation drawing 32A716-02-0G	按照反推装置安装图纸 32A717-02-0G Per T/R installation drawing 32A717-02-0G

注 18/NOTE 18

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